

#396

Xfer to CDDIS

GEOS-C

SATELLITE TO SATELLITE TRACKING

75-027A-06A

ESGP-00088

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1. INTRODUCTION:

The documentation for this data set was originally on paper, kept in NSSDC's Data Set Catalogs (DSCs). The paper documentation in the Data Set Catalogs have been made into digital images, and then collected into a single PDF file for each Data Set Catalog. The inventory information in these DSCs is current as of July 1, 2004. This inventory information is now no longer maintained in the DSCs, but is now managed in the inventory part of the NSSDC information system. The information existing in the DSCs is now not needed for locating the data files, but we did not remove that inventory information.

The offline tape datasets have now been migrated from the original magnetic tape to Archival Information Packages (AIP's).

A prior restoration may have been done on data sets, if a requestor of this data set has questions; they should send an inquiry to the request office to see if additional information exists.

2. ERRATA/CHANGE LOG:

NOTE: Changes are made in a text box, and will show up that way when displayed on screen with a PDF reader.

When printing, special settings may be required to make the text box appear on the printed output.

Version	Date	Person	Page	Description of Change
01				
02				

3 LINKS TO RELEVANT INFORMATION IN THE ONLINE NSSDC
INFORMATION SYSTEM:

<http://nssdc.gsfc.nasa.gov/nmc/>

[NOTE: This link will take you to the main page of the NSSDC Master Catalog. There you will be able to perform searches to find additional information]

4. CATALOG MATERIALS:

- a. Associated Documents To find associated documents you will need to know the document ID number and then click here.
<http://nssdcftp.gsfc.nasa.gov/miscellaneous/documents/>

- b. Core Catalog Materials

GEOS 3

SAT. TO SAT. TRACKING DATA ON TAPE

75-027A-06A ESGP-00088

THIS DATA SET HAS BEEN RESTORED. THERE WAS ORIGINALLY ONE 9-TRACK, 1600 BPI TAPE WRITTEN IN EBCDIC. THERE IS ONE RESTORED TAPE, WRITTEN IN ASCII. THE DR TAPE IS A 3480 CARTRIDGE AND THE DS TAPE IS 9-TRACK, 6250 BPI. THE ORIGINAL TAPE WAS CREATED ON AN IBM 360 COMPUTER AND WAS RESTORED ON AN IBM 9021 COMPUTER. THE DR AND DS NUMBER ALONG WITH THE CORRESPONDING D NUMBER AND TIME SPAN IS AS FOLLOWS:

DR#	DS#	D#	FILES	TIME SPAN
DR005901	DS005901	D025192	16	04/13/75 - 04/18/76

REQ. AGENT
CMP

RAND NO.
RC6860

ACQ. AGENT
JRJ

GEOS-C

SATELLITE TO SATELLITE TRACKING

75-027A-06A

This data set contains 1 tape. The tape is 9-track, 1600 BPI, EBCDIC, with 16 files. The tape format is in the "User's Guide for Satellite to Satellite System Observations and Data Formats."

The 'D' and 'C' numbers along with the time spans is as follows:

<u>D#</u>	<u>C#</u>	<u>TIME SPAN</u>
D-25192	C-19279	04/09/75 - 12/23/75 4/13/75 - 4/18/76

75-027A-064

GEOS-C

**A USER'S GUIDE FOR
SATELLITE TO SATELLITE SYSTEM
OBSERVATIONS AND DATA FORMATS**

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MARCH 1975

GSFC

— GODDARD SPACE FLIGHT CENTER —

GREENBELT, MARYLAND

A USER'S GUIDE FOR SATELLITE TO SATELLITE
SYSTEM OBSERVATIONS AND DATA FORMATS

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March 1975

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A USER'S GUIDE FOR SATELLITE TO SATELLITE
SYSTEM OBSERVATIONS AND DATA FORMATS

J. W. Bryan
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ABSTRACT

This document provides a description of the observations, data formats, and system parameters associated with the Satellite to Satellite Tracking Data Preprocessor Program. The operational Satellite to Satellite Tracking System and Trajectory computation software discussed in this document is the result of a systematic evolution of applied technology which had its beginning in early 1968. The document describes the two types of observations, Range and Doppler, and their functional relation to the satellite states. An appendix provides a complete description of the available (input/output) data formats and the associated system parameters.

SYSTEM DESCRIPTION

The satellite to satellite tracking (SST) system (refs 1, 2 & 3) used to derive the data described in this document consists of; (1) the ground based Application Technology Satellite Ranging (ATSR) system, (modified for satellite to satellite tracking) (2) the wideband communication transponder on the ATS-6 geosynchronous spacecraft and (3) the ranging transponder on the low orbiting satellite. The ATSR system generates the interrogating signals and measures range sum and range sum rate from the returned signals. In the SST configuration the ATSR system generates two C-band radio frequency signals, (1) a beacon carrier used to phase lock the ATS-6 transponder oscillator and (2) a range tone modulated carrier used to measure the range sum and range sum rate. These signals are radiated toward ATS-6 from either Rosman, N.C., Mojave, Calif. or Madrid, Spain ATSR site.

The signals are received by the ATS-6 spacecraft via an earth coverage antenna. Using the phase locked oscillator as a reference frequency, the ATS-6 transponder frequency translates the ranging carrier to S-band. Communications between the ATS-6 and the low orbiting satellite is accomplished using a 9 meter (36 dB gain, 1.4° beamwidth) parabolic antenna on the ATS-6 spacecraft. The low orbiting spacecraft is equipped with either a crystal transponder (NIMBUS) or a phase locked frequency multiplying transponder (GEOS-C and Apollo). The ranging signal received by the low satellite is transponded and radiated back to ATS-6. This return link signal is frequency translated, using the phase locked frequency reference, to C-band in the ATS-6 transponder and radiated by the earth coverage antenna to the ATSR system.

The ATSR system measures the propagation delay by determining the elapsed time between zero crossings of the transmitted and received range tone. This elapsed time is the range sum data output. Since the lowest frequency tone used is 8 Hz, the range measurement is ambiguous for delay times greater than 125 milliseconds. The Doppler is measured using a cycle counting technique. The actual derived Doppler is added to a bias frequency to preserve the sign of the Doppler. The measurement is the time required to accumulate a known number of Doppler plus bias frequency cycles. The Doppler thus determined is the sum of the Doppler offsets on; (1) the 6 GHz ground to ATS-6 link (2) the 2.4GHz ATS-6 to low orbiter link, (3) the 2.25 GHz low orbiter to ATS-6 link and (4) the 4GHz ATS-6 to ground link due to the relative velocities between elements of the individual links.

DESCRIPTION OF THE TRACKING DATA

Range Measurement

The basic observational equation for ranging is simply an expression relating the observed measurement, namely the indicated elapsed time interval, to a so called computed measurement which is a function of the positions of the two satellites at the respective times of signal arrival, the position coordinates of the ground station antenna, and other possible system parameters (e.g. transponder delays, refraction delay). The required relation follows directly from the analysis of the signal geometry which is illustrated in Figure 1.

Let:

τ be the measured or observed time interval for ranging
 t be the recorded observation time based on the ground station clock
 $X_1(u)$, $X_2(u)$, $X_s(u)$ be the geocentric position coordinate vectors for the high satellite, low satellite, and ground station respectively at arbitrary time u .

Then the reference tone which is transmitted from the ground station at time t will reach the high (ATS) satellite at time $t + T_3$, where T_3 represents the transit delay over the link between the ground transmitter and the high (ATS) synchronous satellite. The ATS transponder retransmits the signal after a transponder delay of TD_1 to the low satellite where it arrives at time $t + T_3 + TD_1 + T_1$, where T_1 represents the high to low satellite transit delay. The signal is then retransmitted after a transponder delay TD_2 to the ATS arriving there at time $t + T_3 + TD_1 + T_1 + TD_2 + T_2$, where T_2 represents the low to high satellite transit delay. The signal is finally transponded with delay TD_3 from the ATS to the ground receiver where it arrives at time $t + T_3 + TD_1 + T_1 + TD_2 + T_2 + TD_3 + T_4$, where T_4 represents the signal transit delay over the ATS to ground receiver link. The propagation delays T_3 and T_4 must be corrected for atmospheric refraction, and in the crystal transponder mode the subcarrier receiver delay must be removed from T_4 . Each of the above transit delays is expressible in terms of satellite and ground station position coordinates (for the sake of simplicity, we have neglected other system parameter delays) as follows;

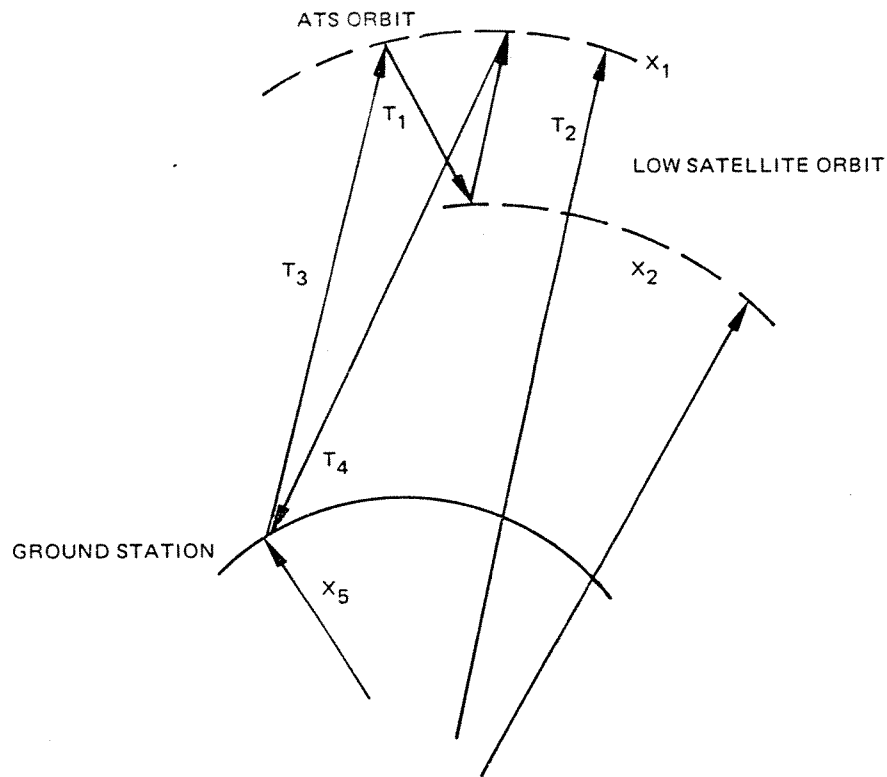
$$T_3 = |X_1(t + T_3) - X_s(t)| / C$$

$$T_1 = |X_2(t + T_3 + T_1) - X_1(t + T_3)| / C$$

(1)

$$T_2 = |X_1(t + T_3 + T_1 + T_2) - X_2(t + T_3 + T_1)| / C$$

$$T_4 = |X_1(t + T_3 + T_1 + T_2) - X_s(t + T_3 + T_1 + T_2 + T_4)| / C$$



X_s is the geocentric station position vector.

X_1 is the ATS position vector

X_2 is the low satellite position vector

$U_3 = T_3$ is the signal delay over ground transmitter to ATS link

$U_1 = T_1$ is the signal transit delay over the ATS to low satellite link

$U_2 = T_2$ is the signal transit delay over the low satellite to ATS link

$U_4 = T_4$ is the signal transit delay over the ATS to ground receiver link

Figure 1. Tracking Geometry

where C is the velocity of signal propagation, and where the expression $|Y|$ means the absolute value of the vector Y.

Finally, then the observed or measured ranging delay τ is expressible in terms of the above as:

$$\text{PLL Mode } \tau = T_1 + T_2 + T_3 + T_4 - n(.125) + \sum_{i=1}^3 TD_i + T_R \quad (2)$$

$$\text{XTAL Mode } \tau = T_1 + T_2 + T_3 + T_4 - n(.125) + T_{sc} + T_R + \sum_{i=1}^3 TD_i$$

where n is the unknown number of 8 Hz (or .125 seconds) range ambiguities.

TD_i is the transponder delay (see Appendix)

T_R is atmospheric correction

T_{sc} is the Subcarrier Receiver Delay (see Appendix)

Doppler Measurement

The Doppler measurement is accomplished by determining the elapsed time interval (T) required to accumulate some fixed number (N), of Doppler plus Bias frequency cycles (Destruct Mode). The Doppler measurement system will later be modified to determine the number (N) of Doppler plus bias frequency cycles for a known period (T) of time (Non destruct Mode). This modification will not change the formulations or processes presented in this document. The relationship between this measured time interval T, and the fixed number of cycles N is given by the expression,

$$N = \phi(t + T) - \phi(t) \quad (3)$$

where $\phi(t)$ represents the phase of the received signal consisting of the Doppler (ref. 3) plus Bias frequency. The entire process undergone by the signal is illustrated in Figure 2 and 3.

The designated path delays are identical with the ones depicted geometrically in Figure 1. Tables I and II list the respective phase of the signal at each reference point of Figures 2 and 3 for a transponder in the XTAL mode of operation, (ref. 2). Similarly, Table III provides the signal phase information for a transponder in the PLL mode, (ref. 3). The required relationship between the observed time delay T, the fixed number N, and the path delays is obtained by substituting the phase of the signal at reference point 6 of Figure 3 into expression (3).

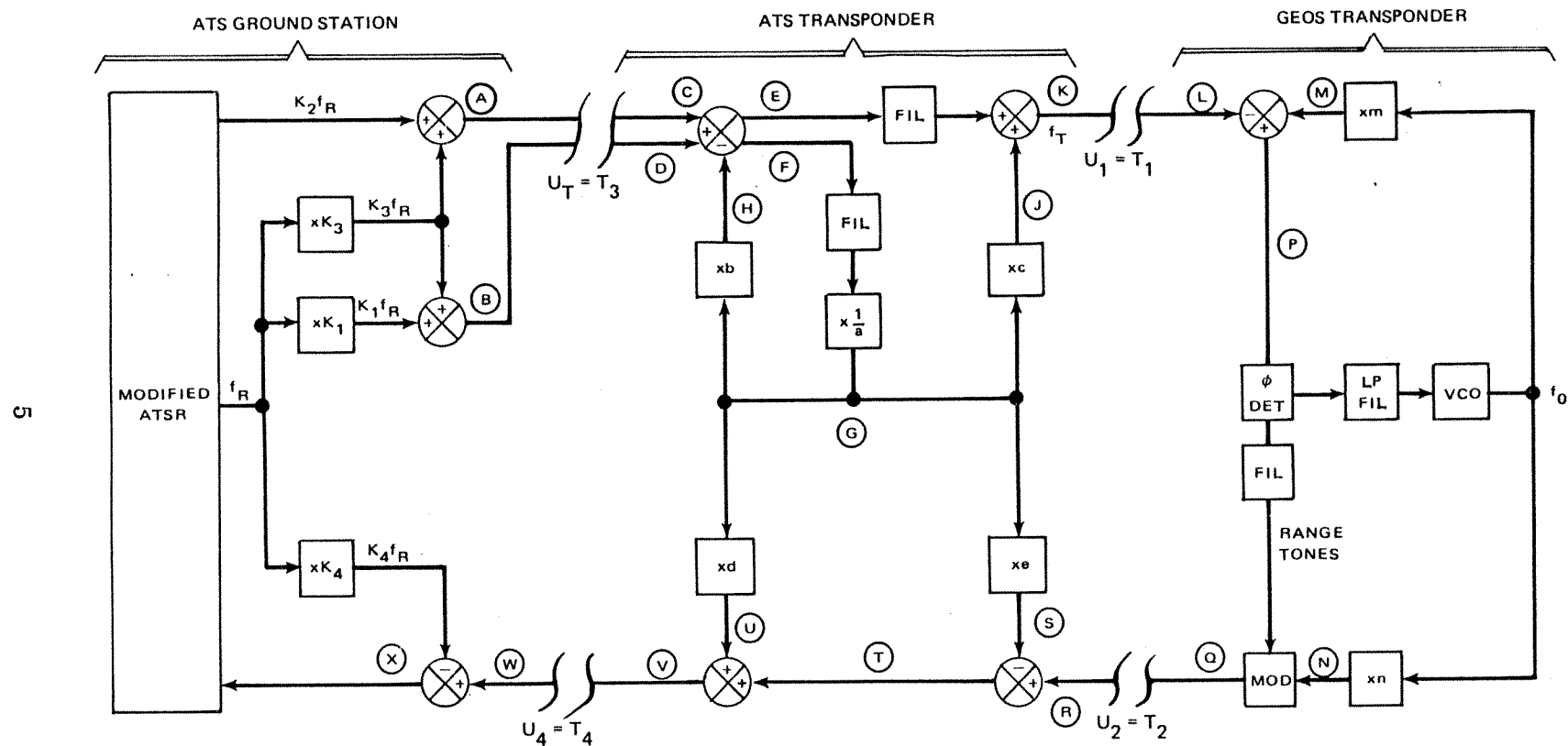
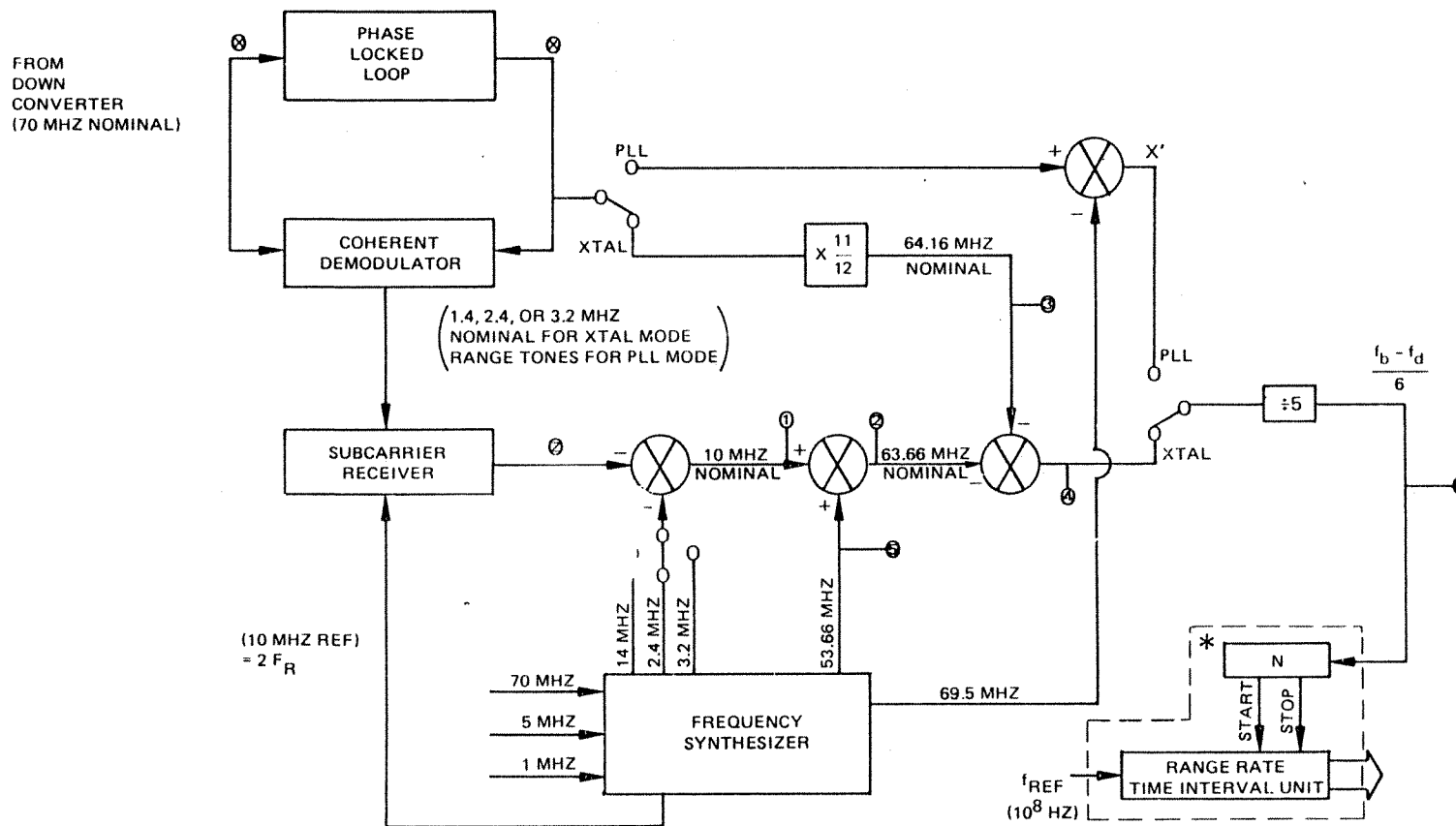


Figure 2. Data Relay Signal Flow Diagram



*TO BE MODIFIED TO INCLUDE CONTINUOUS COUNT SYSTEM

Figure 3. Received Signal Processing and Doppler Extraction Diagram

Table I. Phases for Figure 2 (XTAL Mode)

Fig. 2 Ref. Point	CARRIER PHASE	SUBCARRIER PHASE
A	$(k_2 + k_3) F_R t$	
B	$(k_1 + k_3) F_R t$	
C	$(k_2 + k_3) F_R (t - U_3)$	
D	$(k_1 + k_3) F_R (t - U_3)$	
E	$\left(\frac{1}{a+b}\right) [a(k_2 + k_3) + b(k_2 - k_1)] F_R \cdot (t - U_3)$	
F	$\left(\frac{1}{a+b}\right) [a(k_1 + k_3)] F_R (t - U_3)$	
G	$\left(\frac{1}{a+b}\right) (k_1 + k_3) F_R (t - U_3)$	

Table I (Continued)

Fig. 2 Ref. Point	CARRIER PHASE	SUBCARRIER PHASE
H	$\left(\frac{1}{a+b}\right) [bk_1 + k_3] F_R(t - U_3)$	
J	$\left(\frac{1}{a+b}\right) [c(k_1 + k_3)] F_R(t - U_3)$	
K	$\left(\frac{1}{a+b}\right) [a(k_2 + k_3) + b(k_2 - k_1) + c(k_1 + k_3)]$	
	$F_R \cdot (t - U_3) = F_T(t - U_3)$	
L	$F_T(t - U_3 - U_1)$	
M	$m(F_0 t + \theta)$	
N	$n(F_0 t + \theta)$	

Table I (Continued)

Fig. 2 Ref. Point	CARRIER PHASE	SUBCARRIER PHASE
P	$m(F_0 t + \theta) - F_T(t - U_3 - U_1)$	
Q	$n(F_0 t + \theta)$	$m(F_0 t + \theta) - F_T(t - U_3 - U_1)$
R	$n(F_0(t - U_2) + \theta)$	$m(F_0(t - U_2) + \theta) - F_T(t - U_3 - U_1 - U_2)$
S	$\left(\frac{1}{a+b}\right)[e(k_1 + k_3) F_R(t - U_3)]$	
T	$n(F_0(t - U_2) + \theta) - \left(\frac{1}{a+b}\right)[e(k_1 + k_3)] F_R(t - U_4)$	$m(F_0(t - U_2) + \theta) - F_T(t - U_3 - U_1 - U_2)$
U	$\left(\frac{1}{a+b}\right)[d(k_1 + k_3) F_R(t - U_3)]$	
V	$n(F_0(t - U_2) + \theta) + \left(\frac{d-e}{a+b}\right)(k_1 + k_3) F_R(t - U_3)$	$m(F_0(t - U_2) + \theta) - F_T(t - U_3 - U_1 - U_2)$

Table I (Continued)

Fig. 2 Ref. Point	CARRIER PHASE	SUBCARRIER PHASE
W	$n(F_0(t - U_2 - U_4) + \theta) + \left(\frac{d - e}{a + b}\right)(k_1 + k_3)F_R(t - U_3 - U_4)$	$m(F_0(t - U_2 - U_4) + \theta) - F_T(t - U_3 - U_1 - U_2 - U_4)$
X	$\phi(\omega) - k_4 F_R t$	$\phi(\omega)$

NOTE: For description of system parameters, See Table A-4

Table II. Phases for Figure 3 (XTAL Mode)

Ref Point	Phase of signal at reference point
Y	$(mF_0 - F_T) t - mF_0(U_2 + U_4) + F_T(U_3 + U_1 + U_2 + U_4) + m\theta$
Z	$2F_R t + (mF_0 - F_T) t - mF_0(U_2 + U_4) + F_T(U_3 + U_1 + U_2 + U_4) + m\theta$
1	$2F_R t - mF_0(U_2 + U_4) + F_T(U_3 + U_1 + U_2 + U_4) + m\theta$
2	$\frac{382}{30} F_R t - mF_0(U_2 + U_4) + F_T(U_3 + U_1 + U_2 + U_4) + m\theta$
5	$\frac{322}{30} F_R t$
3	$\frac{m}{n} \left\{ \left[nF_0 + \left(\frac{d-e}{a+b} \right) (k_1 + k_3) F_R - k_4 F_R \right] t - nF_0(U_2 + U_4) \right.$ $\left. - \left(\frac{d-e}{a+b} \right) (k_1 + k_3) (2U_4) F_R + n\theta \right\}$

Table II (Continued)

Ref. Point	Phase of signal at reference point
4	$\left(mF_0 + \frac{m}{n} \left(\frac{d-e}{a+b} \right) (k_1 + k_3) F_R - \frac{m}{n} k_4 F_R - \frac{382}{30} F_R \right) t$ $- \frac{m}{n} \left(\frac{d-e}{a+b} \right) (k_1 + k_3) F_R (U_3 + U_4) - F_T (U_3 + U_1 + U_2 + U_4)$ $= F_B t - \frac{m}{n} \left(\frac{d-e}{a+b} \right) (k_1 + k_3) F_R (U_3 + U_4) - F_T (U_3 + U_1 + U_2 + U_4)$
6	$\frac{F_B}{5} t - \frac{m}{5n} \left(\frac{d-e}{a+b} \right) (k_1 + k_3) F_R (U_3 + U_4) - \frac{F_T}{5} (U_3 + U_1 + U_2 + U_4)$

Table III. Phases for Figures 2 & 3 (PLL Mode)

Fig. 2 Ref. Point	
A - L	See Table I
M	$F_T(t - U_3 - U_1)$
N	$\left(\frac{n}{m}\right) F_T(t - U_3 - U_1)$
Q	$\left(\frac{n}{m}\right) F_T(t - U_3 - U_1)$
R	$\left(\frac{n}{m}\right) F_T(t - U_3 - U_1 - U_2)$
S	See Table I
T	$\left(\frac{n}{m}\right) F_T(t - U_3 - U_1 - U_2) - \frac{1}{(a + b)} (e(k_1 + k_3) F_r(t - U_3))$

Table III (Continued)

Fig. 2 Ref. Point	
U	See Table I
W	$\left(\frac{n}{m}\right) F_T (t - U_3 - U_1 - U_2 - U_4) + \frac{(d - e) (k_1 + k_3)}{(a + b)} F_r (t - U_3 - U_4)$
X	$\left[\left(\frac{n}{m}\right) F_T - \frac{(d - e) (k_1 + k_3)}{(a + b)} F_r - k_4 f_r \right] t - \left(\frac{n}{m}\right) F_T (U_3 + U_1 + U_2 + U_4) - \frac{(d - e) (k_1 + k_3)}{(a + b)} F_r (U_3 + U_4)$
X'	$F_B t - \left(\frac{n}{m}\right) F_T (U_1 + U_3 + U_2 + U_4) - \frac{(d - e) (k_1 + k_3)}{(a + b)} F_r (U_3 + U_4)$
6	$\frac{F_B}{5} t - \left(\frac{n}{m}\right) \frac{F_T}{5} (U_1 + U_3 + U_2 + U_4) - \frac{(d - e) (k_1 + k_3)}{(a + b)} \frac{F_r}{5} (U_3 + U_4)$

NOTE: For description of system parameters see Table A-4.

This yields,

$$N = \frac{f_B}{5} \cdot T + \frac{A_3}{5} (T_4(t+T) - T_4(t) + T_3(t+T) - T_3(t))$$

$$+ \frac{A_4}{5} \cdot \sum_{i=1}^4 (T_i(t+T) - T_i(t)) \quad (4)$$

where,

f_B = bias frequency (nominal value is .5 MHz)

A_3 and A_4 are two additional frequency terms which are expressible in terms of other system parameters, described in Table A-4 as,

$$A_3 = - \frac{(m)}{(n)} \frac{(d-e)}{(a+b)} (k_1 + k_3) f_R \quad (\text{XTAL Mode})$$

$$= - \frac{(d-e)}{(a+b)} (k_1 + k_3) f_R \quad (\text{PLL Mode})$$

$$A_4 = - \frac{f_B}{(a+b)} \{a(k_2 + k_3) + b(k_2 - k_1) + c(k_1 + k_3)\} \quad (\text{XTAL Mode})$$

$$= - \frac{(n)}{(m)} \frac{f_R}{(a+b)} \{a(k_2 + k_3) + b(k_2 - k_1) + c(k_1 + k_3)\} \quad (\text{PLL Mode})$$

Each of the path delays T_i is related to the satellite positions by means of the following nonlinear system of equations,

$$T_4(t) = |X_1(t - T_4(t)) - X_s(t)|/C$$

$$T_2(t) = |X_1(t - T_4(t)) - X_2(t - T_4(t) - T_2(t))|/C$$

(5)

$$T_1(t) = |X_2(t - T_4(t) - T_2(t)) - X_1(t - T_4(t) - T_2(t) - T_1(t))|/C$$

$$T_3(t) = |X_1(t - T_4(t) - T_2(t) - T_1(t)) - X_s(t - T_4(t) - T_2(t) - T_1(t) - T_3(t))|/C$$

where C = velocity of signal propagation. The applicable geometry is illustrated in Figure 1 where system delays have been omitted for simplicity.

In order to properly utilize these satellite to satellite observations, it is necessary that the Orbit Determination Program be capable of synthesizing the equivalent "computed" ranging and doppler observations according to the above non linear observation equations. In the existing Navigation Analysis Program (NAP3) currently utilized at GSFC (ref. 5), these computed observations are synthesized by means of an iterative algorithm, with provisions for applying or solving for a comprehensive set of systematic errors.

It should be noted that the time tag associated with both ranging and doppler observations is referenced to the start of the elapsed range interval time unit counter. This is accounted for in the two separate algorithms used for the synthesis of path delays for ranging and doppler observations as given by means of equation sets (1) and (5) respectively.

The actual value of 'N', the fixed number of doppler plus bias frequency cycles being counted, depends on the data recording rate and is given in Table A-6. Selectable values of 'T' for the Non destruct Mode will be established when the system is modified. These values will be available from GSFC at that time.

REFERENCES

1. Vonbun, F. O. and Mengel, J. T., "Tracking and Communications for Planetary Manned Mission" AIAA Journal of Spacecraft and Rockets, Vol. 5, No. 7, pp. 863-865, July 1968.
2. Schmid, P. E. and Vonbun, F. O., "The ATS-F/NIMBUS-F Tracking and Orbit Determination Experiment" IEEE Intercon, New York, N. Y., March 26-29, 1974.
3. Bryan, J. W., "ATS-F/GEOS-C Satellite to Satellite Tracking Data Processing Considerations," GSFC X-932-74-146, May 1974.
4. Marini, J. W., "Doppler Factors in Satellite to Satellite Tracking," GSFC X-932-74-93, April 1974.
5. Lynn, J. J., "Navigation Analysis Program" GSFC Contract NAS5-11915, Old Dominion Systems Inc., April 1973.

APPENDIX A

Preprocessor Data Formats and System Parameters

A.1 Input Data

The "Tracking Data Preprocessor" can accept tracking data in three formats; the ATSR teletype format, the NAPSIM format, and the GSFC 60-byte format. These data can be input from magnetic tape, disk file or cards. This Appendix contains a detailed description of each of the three types.

A.1.1 ATSR Teletype Format

The characteristic feature of the ATSR teletype data is that two lines are required for each measurement. Figure A-1 contains the present formats of these data for the Destruct and Non-Destruct Doppler count modes. In addition to the data lines, each pass is preceded by a header record which contains identification information. The format of the header record is described in Figure A-2 which contains a sample of an actual satellite tracking.

The two lines contain: two angle measurements, four range rate measurements, four range measurements, a single time tag, a satellite identification code (SAT), a station identification code (STA) and four additional codes C1, C2, C3 and C4.

The angle measurements are ignored by the preprocessor. The range rate measurement is the time required to accumulate a fixed number of bias-plus-doppler frequency cycles (Destruct), or the number of bias plus Doppler frequency cycles accumulated in a measured time period (Non-Destruct). The range measurement is the time delay from signal transmission to receipt. This is an ambiguous measurement in that the clock cycles to zero every 0.125 seconds. Therefore, an approximation of the true range is required to resolve the actual measurement. Such resolution is not performed by the preprocessor.

The time tag of the first doppler and range measurements is that which appears in the first line of data. The time tag is incremented according to the sample rate (C-2) for the remaining three pairs of measurements.

The SAT code contains a three digit number which defines: the relay satellite, the equipment mode and the target object. Table A-1 contains a matrix which identifies the elements of SAT. Referring to this table, consider the three digit code 642. Then;

- 6 = Relay Satellite ID = ATS-6.
- 4 = Equipment Mode = Crystal on ground, and
- 2 = Target object = Santiago, Chile.

Also, note from this table that for SAT ≥ 640 , the target object is some ground based transponder. The raw tracking data contains the two digit identification, STA, of the transmitting station. For those data which are relayed through a ground transponder, the Preprocessor constructs an identification code for the transponder. Table A-2 contains the station numbers transponder type and transponder delay valid at the time of this document. The C1 parameter is not used by the preprocessor. The C2 and C3 parameters are used to extract information necessary in the synthesis of the doppler measurement. Table A-3 lists the locations of the stations used thus far in the various SST modes.

The C3 parameter identifies the uplink and downlink frequencies used in the satellite relay tracking. There are four terms of the doppler equation which are dependent on these frequencies. Table A-4 contains for each possible value of C3: the uplink and downlink frequencies, and the values of the four terms. This table also contains the values of nine additional terms which are independent of frequency but are required for the doppler synthesis. These terms are used in the calculation of the doppler constants:

$$A_3 = -\frac{m(d - e)}{n(a + b)} (k_1 + k_3) f_r (\text{XTAL Mode})$$

$$A_4 = -\frac{f_r}{a + b} \{a(k_2 + k_3) + b(k_2 - k_1) + c(k_1 + k_3)\} (\text{XTAL Mode})$$

For PLL mode, A_3 and A_4 are multiplied by the ratio n/m .

There are two additional constants required for the doppler synthesis which are obtained from the tracking data. These are:

f_b = frequency bias and

N = cycle count

The bias frequency consists of a nominal value plus a differential which varies according to the transmitting station, equipment mode, target object and operating frequency.

$$f_b = 5.D5 + df \quad (\text{Hertz})$$

where: df = bias frequency differential.

The possible values of df for the Rosman and Mojave Mobile transmitters are contained in Table A-5.

The remaining constant N for the Destruct mode of Doppler counting, is dependent on the sample rate which is identified via the C4 parameter. Table A-6 contains the possible values for this parameter. All of the tables discussed in this Appendix are contained in the Preprocessor. The doppler constants, which are dependent on the values of these tables, are computed and output. A more complete description of the mathematics, thus employed, is contained in the main section of this document.

A.1.2 NAPSIM Format

The NAPSIM format is that used by the "Navigation Analysis Program", NAP. This format consists of a header record for each pass, one record which contains the measurements at each time point and a terminal record. Each record contains eight 8-byte binary words.

A.1.2.1 The Leader Record Contains:

- Word 1 = 0.D0
 - 2 = Arc Number = 1.D0 for all Preprocessed data.
 - 3 = Station Number = Two digits station identification number, see Table A-2
 - 4 = Pass number = a serial identification associated with each pass for each station.
 - 5 = Target Object
 - = -2 if sidetone or coherent ATS tracking
 - = 0 if satellite to satellite tracking otherwise, the two digit remote transponder identification of Table A-2.
 - 6 = Name of transmitting site
 - 7 = Name of target object
 - 8 = Byte 1 = Equipment Mode
 - 2 = Target Object
 - 3 = C1
 - 4 = C2
 - 5 = C3
 - 6 - 8 = spares
- (See section A.1.1 for further identification)

A.1.2.2 Each Data Record Contains:

Word 1 = Decimal days since January 0, 1950
2 = Seconds of current day
3 = Range measurement
4 = Doppler measurement
5 - 8 = 0.D0

A.1.2.3 The Final Record of Each File Contains:

Word 1 = -1.D0
2 - 8 = 0.D0

A.1.3 GSFC 60-Byte Format

The GSFC 60-byte general data format used by GSFC Code 500 can also be read by the preprocessor. Table A-7 contains a general description of each of the thirteen words comprising this data type. Words 1 and 2 contain identification codes of the receiving and transmitting sites respectively. Table A-8 correlates these codes with the actual sites.

For satellite-to-satellite tracking, word 12 will contain the integer 13, word 6: the doppler count in seconds, word 7: the range sum in kilometers, and word 8: the doppler in hertz.

Word 9 contains additional information extracted from the raw data. Table A-9 summarizes the contents of this word. Each of the variables contained in word 9 is more fully described in Section A.1.1 ATSR teletype format.

A.2 Output Data

The Preprocessor can produce data tapes in the NAPSIM, 60-byte or MTAUF formats. At the user's option, tapes in any or all of the three formats can be output in a single run. The NAPSIM and 60-byte tapes can contain either all of the observations or compacted observations. The compacted observations are obtained from an n^{th} degree polynomial over N data points. The degree of the polynomial and the number of measurements to be compacted is under user control. The data formats are not altered if the data is compacted.

The formats of the NAPSIM and 60-byte tapes are fully described in the Input Data portion of this appendix. The remaining format, the MTAUF, will now be described.

A.2.1 MTAUF Format

The Mission Trajectory and Analysis Unified Format (MTAUF) is an alternate NAP input format. This format is available only for compacted data. The degree of the polynomial and the number of points to be compacted to a single observation are under the user's control. Table A-10 contains the format of this data type. The tape produced by the preprocessor will contain the following:

Word 1 = Two digit ID of transmitting station
2 = 9
3 = Days
4 = Seconds
5 - 8 = 0.D0
9 = Elapsed time for ranging
10 = Elapsed time of N cycles of doppler plus bias
11 = 0.D0

The primary disadvantage to this data format is the absence of: the ID of the receiving site, the equipment mode, the target object, the constant C2 and the constant C3. All of this information is essential for proper processing. Therefore, this information must be obtained from other Preprocessor outputs prior to data reduction.

Table A-1
ATSR Satellite ID, SAT

A-1.1 ATS Identification

SAT	ATS Series
0xx	ATS-1 VHF
1xx	ATS-1
2xx	ATS-3 VHF
3xx	ATS-3
5xx	ATS-5
6xx	ATS-6

A-1.2 Satellite To Satellite Identification

Equipment Mode	Target Object							
	= 0	= 1	= 2	= 3	= 4	= 5	= 6	= 7
0 = Sidetone	ATS-6			Boresite				
1 = Coherent	Nimbus	GEOS	ASTP	Boresite				
2 = PLL Satellite	GEOS	ASTP						
3 = XTAL Satellite	Nimbus							
4 = XTAL Ground	Rosman	Mojave	Santiago	Mobile	NTTF	Houston	Madrid	Ascension
5 = PLL Ground	Rosman	Mojave	Santiago	Mobile	NTTF	ASTP Houston	ASTP Madrid	

Table A-2
Station Identification

Station Name	Type	ID No.
Rosman	Xmitter	58
Mojave Mobile	Xmitter	66
Madrid Mobile	Xmitter	99
Rosman	Xtal Xpndr	40
Rosman	PLL Xpndr	50
Mojave	Xtal Xpndr	41
Mojave	PLL Xpndr	51
Santiago	Xtal Xpndr	42
Santiago	PLL Xpndr	52
Boresite	Xtal Xpndr	43
Boresite	PLL Xpndr	53
NTTF	Xtal Xpndr	44
NTTF	PLL Xpndr	54
Port Loc	Xtal Xpndr	45
Port Loc	PLL Xpndr	55
Ascension Island	Xtal Xpdr	
Johannesburg	Xtal Xpdr	

Table A-2.1
Summary of Equipment Delays (Nanoseconds)

Station	Type	Total Delay (including cables)
Rosman	ATSR PLL Mode	0
Rosman	ATSR XTAL Mode	780
Madrid	ATSR PLL Mode	0
Madrid	ATSR XTAL Mode	780
Rosman	XTAL XPDR	1719
Mojave	XTAL XPDR	1830
Santiago	XTAL XPDR	1791
Madrid	XTAL XPDR	1625*
Madrid	Apollo XPDR Simulator	to be determined**
Johannesburg	XTAL XPDR	1793
Ascension	XTAL XPDR	1791
NTTF	XTAL XPDR	2731
NTTF	PLL XPDR	1605
ATS-6	C-band to S-band	170
ATS-6	S-band to C-band	190
GEOS-3	PLL XPDR	1580
NIMBUS-F	XTAL XPDR	2541

*Cable Delay not available as of publication date

**Apollo XPDR simulator delay will be measured immediately prior to
ASTP mission

Table A-3
Station Locations

Station Name	Geodetic Coordinates		
	Latitude	Longitude	GEOID Height
Rosman (ATSR)	35° 11' 55".6	277° 07' 27".5	824 m
Mojave (ATSR)	35° 19' 53".9	243° 06' 47".7	886 m
Madrid (ATSR)	40° 27' 19".7	355° 49' 53".9	808 m
Ahmedabad (ATSR)	23° 01' 19".8	72° 30' 54".9	250 m
Ascension*	-07° 57' 19".	345° 40' 20".7	528 m
Santiago*	-33° 09' 04".9	289° 19' 54".2	703 m
NTTF*	38° 59' 53".6	283° 09' 27".8	-1 m
Johannesburg*	-25° 53' 11".0	27° 42' 26".9	1542 m

*Transponder locations are approximated

Table A-4
Doppler Synthesis Terms

A-4.1 Frequency Dependent Terms

C3	Frequency (MHz) Uplink/Downlink	k_3	k_4	b	d
0	NOT ATS-6	-	-	-	-
1	5950/3750 (XTAL)	1173.5700	736.6	232	144
	(PLL)	1174.8225	755.4	232	144
2	5950/3950 (XTAL)	1173.5700	776.6	232	152
	(PLL)	1174.8225	775.9	232	152
3	5950/4150 (XTAL)	1173.5700	816.6	232	160
	(PLL)	1174.8225	815.9	232	160
4	6150/3750 (XTAL)	1213.5700	736.6	240	144
	(PLL)	1214.8225	735.4	240	144
5	6150/3950 (XTAL)	1213.5700	776.6	240	152
	(PLL)	1214.8225	775.4	240	152
6	6150/4150 (XTAL)	1213.5700	816.6	240	160
	(PLL)	1214.8225	815.4	240	160
7	6350/3750 (XTAL)	1253.5700	736.6	248	144
	(PLL)	1254.8225	735.4	248	144
8	6350/3950 (XTAL)	1253.5700	776.6	248	152
	(PLL)	1254.8225	775.4	248	152
9	6350/4150 (XTAL)	1253.5700	816.6	248	160
	(PLL)	1254.8225	815.4	248	160

A-4.2 Frequency Independent Terms

$$f_r = 5.0 \text{ MHz}$$

$$f_o = 37.55 \text{ MHz}$$

$$m = 55 \text{ (XTAL)}$$

$$n = 60 \text{ (XTAL)}$$

$$= 221 \text{ (PLL)}$$

$$= 240 \text{ (PLL)}$$

$$a = 6$$

$$k_2 = 14$$

$$c = 77$$

$$e = 84$$

$$k_1 = 16.43 \text{ (XTAL)}$$

$$k_1 = 15.1775 \text{ (PLL)}$$

Table A-5

A-5.0.0 Doppler Bias Frequencies

A-5.1.0 Transmitter = Rosman

A-5.1.1 Equipment Mode = Crystal

FREQ.	NIMBUS	FREQ.	NIMBUS
1	-14.29557517	6	22.39718083
2	-28.94252179	7	-14.31895117
3	22.41053164	8	-28.96714011
4	-14.30764303	9	22.38467101
5	-28.95523113		

A-5.1.2 Equipment Mode = Phase Locked Loop

FREQ.	GEOS	ASTP	
1	32.30279344	31.99999907	
2	16.31892826	15.99999888	
3	0.33506251	-0.00000131	
4	32.29294646	31.99999878	
5	16.30855633	15.99999860	
6	0.32416619	-0.00000159	
7	32.28371992	31.99999907	
8	16.29883837	15.99999888	
9	0.31395626	-0.00000131	

A-5.1.3 Equipment Mode = Coherent

FREQ.	NIMBUS	GEOS	ASTP
1	8.40336111	20.30252102	31.99999964
2	-7.57512669	4.31865556	15.99999945
3	-23.55361477	-11.66521105	-0.00000074
4	8.39024364	20.29268285	31.99999964
5	-7.58894362	4.30829272	15.99999945
6	-23.56813116	-11.67609855	-0.00000074
7	8.37795238	20.28346455	31.99999964
8	-7.60189027	4.29858272	15.99999945
9	-23.58173350	-11.68630024	-0.00000074

A-5.2.0 Transmitter = Mojave Mobile (Hybrid)

A-5.2.1 Equipment Mode = Crystal

FREQ.	NIMBUS	FREQ.	NIMBUS
1	-12.83333336	6	7.33333320
2	55.00000054	7	-12.83333364
3	7.33333320	8	55.00000026
4	-12.83333336	9	7.33333292
5	55.00000054		

A-5.2.2 Equipment Mode = Phase Locked Loop

FREQ.	GEOS	ASTP	
1	43.08224459	51.99999930	
2	120.44358994	-0.00000131	
3	71.80493395	-52.00000163	
4	41.03078948	51.99999902	
5	118.28282294	-0.00000159	
6	69.53485507	-52.00000192	
7	39.10855992	51.99999930	
8	116.25816694	-0.00000131	
9	67.40777320	-52.00000163	

A-5.2.3 Equipment Mode = Coherent

FREQ.	NIMBUS	GEOS	ASTP
1	-25.99999948	85.02520996	-2.00000062
2	48.00000096	36.38655426	-54.00000009
3	-4.00000022	113.74789847	19.99999864
4	-25.99999948	82.97560993	-2.00000062
5	48.00000096	34.22764206	-54.00000009
6	-4.00000022	111.47967439	19.99999949
7	-26.00000005	81.05511807	-2.00000034
8	48.00000039	32.20472405	-54.00000009
9	-4.00000022	109.35433079	19.99999949

A-5.3.0 Identification of FREQ. Codes

Frequency	Uplink (MHz)	Downlink (MHz)
1	5950	3750
2	5950	3950
3	5950	4150
4	6150	3750
5	6150	3950
6	6150	4150
7	6350	3750
8	6350	3950
9	6350	4150

Table A-6
ATSR Sample Rate

C4	N	Sample Rate
0	51192	1 / sec
1	25596	2 / sec
2	12798	4 / sec
3	6399	8 / sec
4	511920	6 / min

Table A-7
Word Definition

Word	Content	Byte	Type
1	Station ID and equipment type (downlink)	1-2	I*2
2	Station ID and equipment type (uplink)	3-4	I*2
3	Year of observation	5-6	I*2
4	Day of year of observation	7-8	I*2
5	Frame time of observation	9-16	R*8
6	Doppler count interval	17-24	R*8
7	Observation 1	25-32	R*8
8	Observation 2	33-40	R*8
9	Data flags	41-44	I*4
10	Data corrections	45-48	I*4
11	SIC/NORAD ID	49-52	I*4
12	Data type	53-54	I*2
13	Spares	55-60	—

Table A-8
ATSR Site Identification Codes, 60-Byte Data

60-Byte Index Number	Station Name	NAPSIM Station Code
55	Rosman	58
56	Kashima	68
57	Mobile	66 = Mojave Mobile 99 = Madrid Mobile

Table A-9
Word 9, 60-Byte Data

Byte	Bits	Value	Contents
1	1-4		Equipment Mode ~ See Table A-1.2
	5-8		Target Object ~ See Table A-1.2
2	1-4		C1 parameter ~ Not used by Preprocessor
	5-8		Q/D Flag ~ Not used by Preprocessor
3	1-5	0	Set to zero
	6-8		C2 parameter ~ See Table A-6
4	1-4 5-8		Satellite ID character 1, See Table A-1.1
			C3 parameter: Frequency indicator
		0	ATS-1, 3 and 5
		1	5950 and 3750 MHz
		2	5950 and 3950 MHz
		3	5950 and 4150 MHz
		4	6150 and 3750 MHz
		5	6150 and 3750 MHz
		6	6150 and 4150 MHz
		7	6350 and 3750 MHz
		8	6350 and 3950 MHz
		9	6350 and 4150 MHz

Table A-10
MTAUF Format

Data Word	Type	Description
1	I*4	Station Identification
2	I*4	Number of R*8 words to follow
3	R*8	Julian Days since 1950
4	R*8	Seconds of day
5	R*8	X-angle
6	R*8	Y-angle
7	R*8	Azimuth
8	R*8	Elevation
9	R*8	Range
10	R*8	Range Rate
11	R*8	Δt associated with average range rate

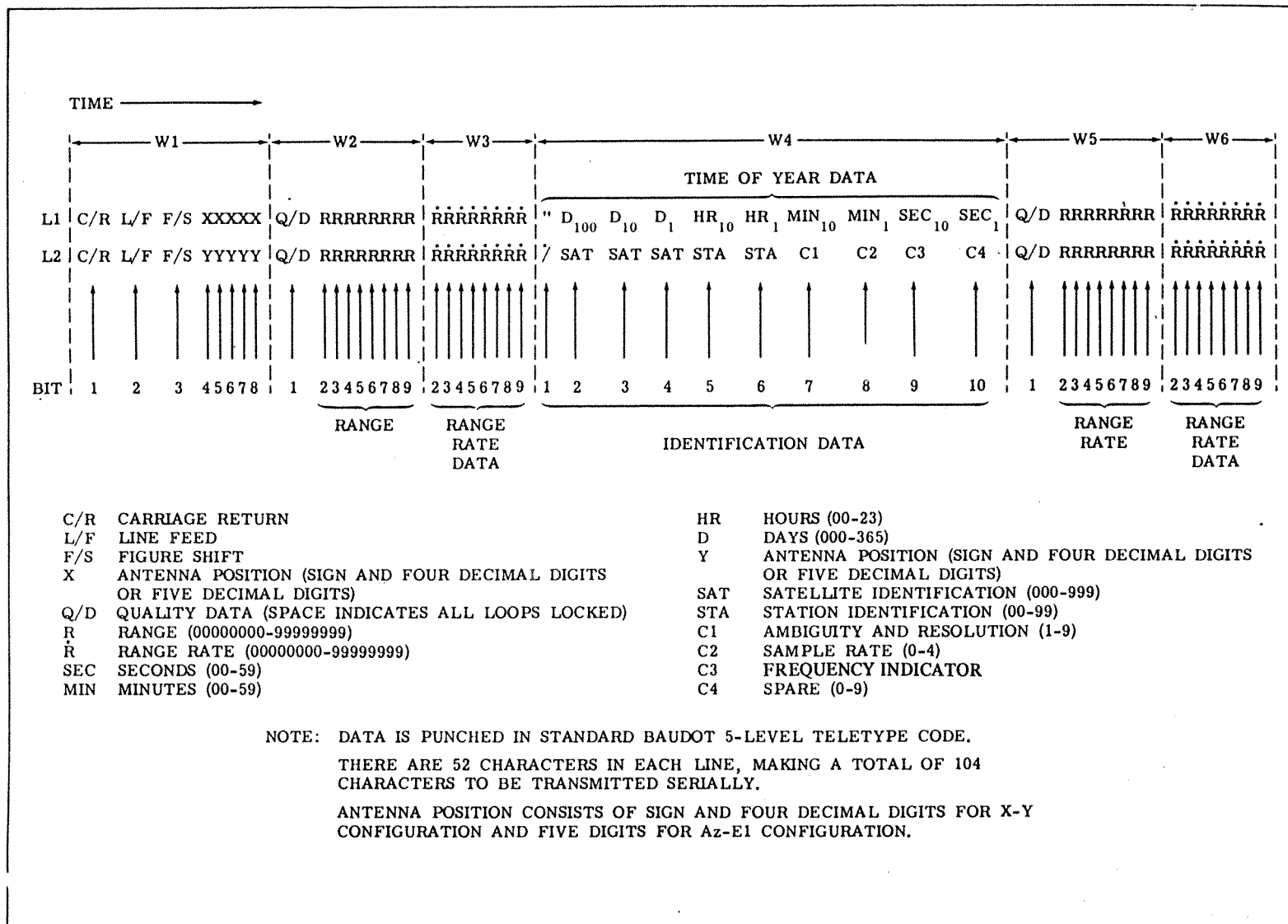
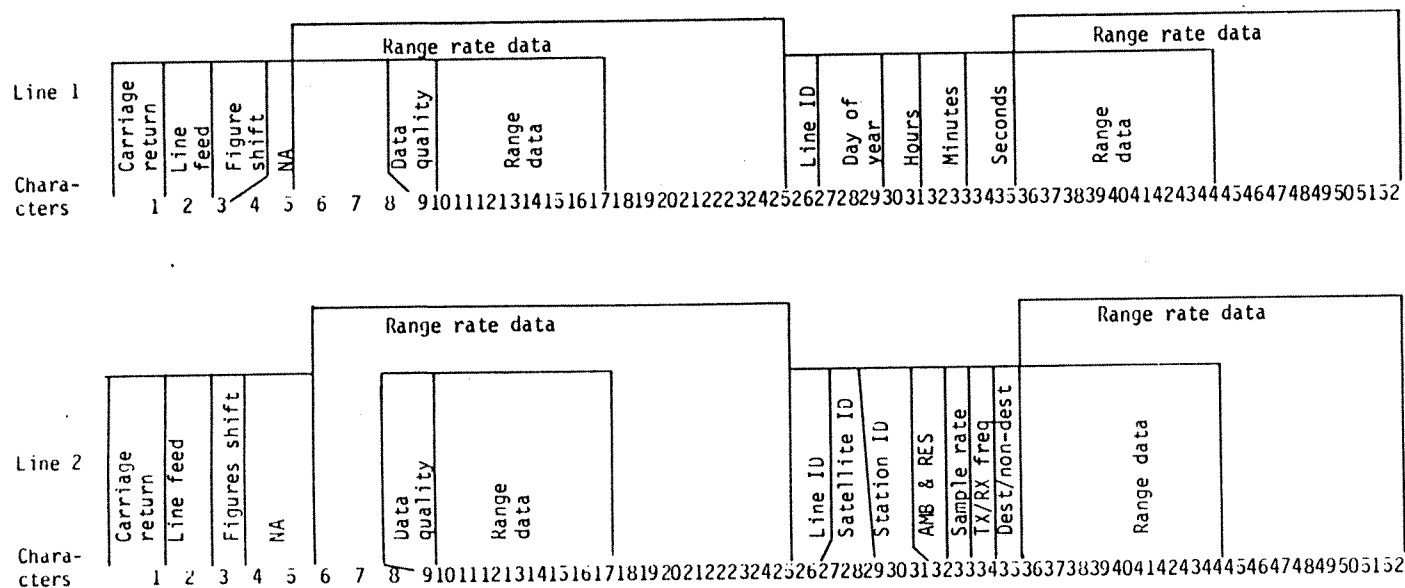


Figure A-1.1



Legend

1. Characters 4 and 5, lines 1 and 2 are not applicable and will indicate (99).
2. A space in the data quality indicates all loops locked.
3. Range data (00000000-99999999).
4. Range rate data (0000000000-9999999999).
5. Line 1 ID ("), line 2 ID (/).
6. Ambiguity and resolution (1-9).
7. Sample rate 6/min will indicate (0).
8. Transmit/receive base frequencies (1-9).
9. Destruct mode (0), Non-destruct (1).

Note

1. Data is punched in Baudot 5-level teletype code.
2. There are 52 characters in each line making a total of 104 characters to be transmitted serially.

Figure A-1.2. Tracking Data Format for Non-destruct Mode

	Data ID Code	Inter- national Desig- nator	Date of Tracking
Header Record	..ATSR	7403901	750112

First	Range	Doppler	Day	Hr	Mn	Sec	Range	Doppler
Data Line	3377	'0014441451196626	"0121101	'00'			'0014441251196616'	

Second	Range	Doppler	SAT	STA	SSC	SSC	Range	Doppler
Data Line	-3762	0014441051196647	/600473050				0014440951196653	

Sample Pass	..ATSR	7403901	740606
-1984	1236218951174105	157183331	1236219651174577
-3903	1236220351174141	/600583040	1236221051174252
-1984	1236221751174569	157183335	1236222351174105
-3903	1236223051174247	/600583040	1236223751174613
-1982	1236224451174097	157183339	1236225151174275
-3904	1236225851174321	/600583040	1236226451174011
-1983	1236227251174391	157183343	1236227951174521
-3904	1236228551173956	/600583040	1236229351174376
-1984	1236230051174512	157183347	1236230751173982
-3903	1236231451174401	/600583040	1236232051174468
-1983	1236232751173990	157183351	1236233451174518
-3905	1236234151174362	/600583040	1236234851174147
-1983	1236235551174631	157183355	1236236151174313
-3905	1236236951174067	/600583040	1236237651174530
-1984	1236238251174165	157183359	1236238951174263
-3903	1236239651174451	/600583040	1236240351174090
-1983	1236241051174223	157183363	1236241751174543
-3904	1236242451173999	/600583040	1236243051174210
-1983	1236243751174564	157183367	1236244551174025
-3905	1236245251174379	/600583040	1236245851174472
-1985	1236246551174039	157183371	1236247251174467
-3904	1236247851174486	/600583040	1236248551174066
-1984	1236249251174435	157183375	1236249951174475
-3904	1236250651174066	/600583040	1236251351174430
-1982	1236252151174305	157183379	1236252751174115
-3905	1236253451174539	/600583040	1236254151174115
-1984	1236254851174095	157183383	1236255551174551
-3905	1236256251174207	/600583040	1236256951174177
-1985	1236257651174489	157183387	1236258351174070
-3904	1236259051174132	/600583040	1236259751174604
-1984	1236260451174059	157183391	1236261151174265
-3904	1236261751174539	/600583040	1236262451173908
-1983	1236263151174414	157183395	1236263851174528
-3905	1236264651174022	/600583040	1236265251174349
-1984	1236265951174431	157183399	1236266651174110
-3905	1236267351174378	/600583040	1236267951174367
-1985	1236268751174034	157183403	1236269451174442
-3905	1236270051174301	/600583040	1236270851174032
-1983	1236271451174613	157183407	1236272151174248
-3904	1236272851174081	/600583040	1236273551174517

Figure A-2. Annotated Sample of ATSR Non-Destruct Teletype Data

IEF285I SYS1,PL1LIB KEPT
IEF285I VOL SER NOS= K3SYS1.
IEF285I SYS1,FORTSP KEPT
IEF285I VOL SER NOS= K3SYS2.
IEF285I SYS2,CDBLIB KEPT
IEF285I VOL SER NOS= M2JOB0.
IEF285I SYS1,ALGLIB KEPT
IEF285I VOL SER NOS= K3SYS1.
IEF285I SYS77159,T171726,RV000,YZRWPMA2,LODMOD PASSED
IEF285I VOL SER NOS= K3SCR5.
IEF285I SYS77159,T171726,SV000,YZRWPMA2,R0000457 SYSOUT
IEF285I VOL SER NOS= K3SCR4.
IEF285I SYS77159,T171726,SV000,YZRWPMA2,R0000458 SYSOUT
IEF285I VOL SER NOS= K3SCR5.
IEF285I SYS77159,T171726,RV000,YZRWPMA2,R0000459 DELETED
IEF285I VOL SER NOS= K3SCR3.
IEF285I SYS77159,T171726,RV000,YZRWPMA2,S0000460 SYSIN
IEF285I VOL SER NOS= K3SCR3.
IEF285I SYS77159,T171726,RV000,YZRWPMA2,S0000460 DELETED
IEF285I VOL SER NOS= K3SCR3.
IEF373I STEP /LINK / START 77159,2003
IEF374I STEP /LINK / STOP 77159,2004 CPU 0MIN 00.99SEC MAIN 130K LCS 0K
- STEP 01 - RETURN CODE = 0000 STEP TIME = .20 MINS=(CPU= .01,IO= .19)

IO IN SECS. DISK= 11.87,DRUM= .00,TAPE= .00,CEL= .00,OTHR= .18
- SURCHARGES=(DRIVES ALOC=000,TAPE MOUNTS=000,CORE=000,PAPER=000,PRIORITY=00000)SECS. TOTAL STP TIME= .20 MINS. -

XXGO EXEC PGM=*,LINK,SYSMOD,COND=(4,LT),REGION=70K 00000240
XXFT05F001 DD DDNAME=DATA5 00000250
XXFT06F001 DD SYSOUT=EQOUT,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=8BLKSIZE) 00000260
IEF653I SUBSTITUTION JCL - SYSOUT=A,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=7265) 00000270
XXFT07F001 DD SYSOUT=B,DCB=(RECFM=FB,BLKSIZE=7280,LRECL=80) 00000280
XXSYSPRINT DD SYSOUT=EQOUT,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=8BLKSIZE), 00000290
IEF653I SUBSTITUTION JCL - SYSOUT=A,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=7265),
XX SPACE=(CYL,(0,1)),UNIT=(DISK,3) 00000300
XXSYSUDUMP DD SYSOUT=A,SPACE=(CYL,(0,5)) 00000300
//GO.FT08F001 DD UNIT=(9TRACK,,DEFER),LABEL=(,BLP,,IN),DISP=OLD,
// DCB=(BLKSIZE=32000,RECFM=U,DEN=3),
// VOL=SER=JJ0097
//GO.DATA5 DD *

IEF236I ALLOC. FOR YZRWPMA2 GO
IEF237I 333 ALLOCATED TO PGM=*,DD
IEF237I 231 ALLOCATED TO FT05F001
IEF237I 333 ALLOCATED TO FT06F001
IEF237I 333 ALLOCATED TO FT07F001
IEF237I 331 ALLOCATED TO SYSPRINT
IEF237I 231 ALLOCATED TO SYSPRINT
IEF237I 232 ALLOCATED TO SYSPRINT
IEF237I 333 ALLOCATED TO SYSUDUMP
IEF237I 0D5 ALLOCATED TO FT08F001

RECORD 1 OF FILE 1
LENGTH = 1800 BYTES

ATSR SAT ED 750413 0010841751202801103080035
4B4BC1E3 F2D940F7 E4E0E3E9 E0E140F7 F5F0F4F1 F3404040 40404040 40404040 40404040 40404040
40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040
F1F0F8F4 F7F7F5F1 F2F0F2F0 F0F175F1 F0F3F0F8 F0F0F3F0 40F0F0F1 F0F8F3F3 F4F5F1F2 F0F1F9F0
F7404040 40404040 40404040 40404040 40404040 40404040 60F3F6F9 F640F0F0 F1F0F8F3 F3F0F5F1
F2F0F1F8 F9F361F6 F0F0F4F7 F3F0F5F0 40F0F0F1 F0F8F3F2 F6F5F1F2 F0F1F9F2 F3404040 40404040
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RECORD 375 OF FILE 1
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375 RECORDS IN FILE 1 OF TAPE

RECORD 1 OF FILE 16
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RECORD 148 OF FILE 16
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F1F9F2F0	F4F4F2F0	F2F0F0F8	F5F0F2F4	F0F7F0F6	F2F4F4F3	F8404040	40404040	40404040	40404040
40404040	40404040	F3F9F0F4	F440F0F0	F8F5F0F2	F3F5F7F0	F6F7F8F0	F5F661F6	F1F1F9F9	F2F4F5F1
F6F0F0F8	F5F0F2F3	F0F7F0F7	F3F1F6F8	F7404040	40404040	40404040	40404040	40404040	40404040
F3F9F0F4	F840F0F0	F8F5F0F2	F2F5F7F0	F7F8F5F3	F4F97FF1	F1F9F2F0	F4F5F0F0	F0F0F0F8	F5F0F2F1
F9F7F0F8	F3F9F0F2	F6404040	40404040	40404040	40404040	40404040	40404040	F3F9F0F5	F240F0F0
F8F5F0F2	F1F5F7F0	F8F9F2F7	F3F161F6	F1F1F9F9	F2F4F5F1	F4F0F0F8	F5F0F2F1	F0F7F0F9	F4F6F4F5
F6404040	40404040	40404040	40404040	40404040	40404040	F3F9F0F5	F640F0F0	F8F5F0F2	F0F4F7F1
F0F0F0F2	F0F67FF1	F1F9F2F0	F4F5F4F0	F8F0F0F8	F5F0F1F9	F9F7F1F0	F5F3F9F7	F9404040	40404040
40404040	40404040	40404040	40404040	F3F9F0F6	F040F0F0	F8F5F0F1	F9F5F7F1	F1F0F7F7	F6F861F6
F1F1F9F9	F2F4F5F1	F2F0F0F8	F5F0F1F9	F0F7F1F1	F6F1F5F8	F3404040	40404040	40404040	40404040
40404040	40404040								

148 RECORDS IN FILE 16 OF TAPE

IEF142I	-	STEP WAS EXECUTED - COND CODE 0000	
IEF285I	SYS77159.T171726.RV000.YZRWPMA2.LODMOD	PASSED	
IEF285I	VOL SER NOS= K3SCR5.		
IEF285I	SYS77159.T171726.RV000.YZRWPMA2.S0000466	SYSIN	
IEF285I	VOL SER NOS= K3SCR2.		
IEF285I	SYS77159.T171726.RV000.YZRWPMA2.S0000466	DELETED	
IEF285I	VOL SER NOS= K3SCR2.		
IEF285I	SYS77159.T171726.SV000.YZRWPMA2.R0000461	SYSCUT	
IEF285I	VOL SER NOS= K3SCR5.		
IEF285I	SYS77159.T171726.SV000.YZRWPMA2.R0000462	DELETED	
IEF285I	VOL SER NOS= K3SCR5.		
IEF285I	SYS77159.T171726.SV000.YZRWPMA2.R0000463	DELETED	
IEF285I	VOL SER NOS= K3SCR4.		
IEF285I	SYS77159.T171726.SV000.YZRWPMA2.R0000464	DELETED	
IEF285I	VOL SER NOS= K3SCR5.		
IEF285I	SYS77159.T171726.RV000.YZRWPMA2.R0000465	KEPT	
IEF285I	VOL SER NOS= JJ0097.		

IEF285I SYS1,PL1LIR KEPT
IEF285I VOL SER NOS= K3SYS1.
IEF285I SYS1,FORTSSP KEPT
IEF285I VOL SER NOS= K3SYS2.
IEF285I SYS2,CORLIR KEPT
IEF285I VOL SER NOS= M2JOB0.
IEF285I SYS1,ALGLIR KEPT
IEF285I VOL SER NOS= K3SYS1.
IEF285I SYS77160,T103906,RV000,YZRWPMA2,LOD MOD PASSED
IEF285I VOL SER NOS= K3SCR2.
IEF285I SYS77160,T103906,SV000,YZRWPMA2,R0000285 SYSOUT
IEF285I VOL SER NOS= K3SCR3.
IEF285I SYS77160,T103906,SV000,YZRWPMA2,R0000286 SYSOUT
IEF285I VOL SER NOS= K3SCR4.
IEF285I SYS77160,T103906,RV000,YZRWPMA2,R0000287 DELETED
IEF285I VOL SER NOS= K3SCR4.
IEF285I SYS77160,T103906,RV000,YZRWPMA2,S0000288 SYSIN
IEF285I VOL SER NOS= K3SCR2.
IEF285I SYS77160,T103906,RV000,YZRWPMA2,S0000288 DELETED
IEF285I VOL SER NOS= K3SCR2.

IEF373I STEP /LINK / START 77160.1542
IEF374I STEP /LINK / STOP 77160.1543 CPU 0MIN 01.04SEC MAIN 130K LCS OK
- STEP 01 - RETURN CODE = 0000 STEP TIME = .20 MINS=(CPU=.01,I/O=.19)

IO IN SECS. DISK= 11.77,DRUM= .00,TAPE= .00,CELL= .00,OTHR=.18
- SURCHARGES=(DRIVES'ALOC=000,TAPE MOUNTS=000,CCRE=000,PAPER=000,PRIORITY=0000)SECS. TOTAL STP TIME= .20 MINS. -

XXGO EXEC PGM=*,LINK,SYSLMOD,CCND=(4,LT),REGION=70K 00000240
XXFT05F001 DD DDNAME=DATA5 00000250
XXFT06F001 DD SYSOUT=8OUT,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=6BLKSIZE) 00000260
IEF653I SUBSTITUTION JCL - SYSOUT=A,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=7265)
XXFT07F001 DD SYSOUT=B,DCB=(RECFM=FB,BLKSIZE=7265,LRECL=30) 00000270
XXSYSPRINT DD SYSOUT=8OUT,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=6BLKSIZE), 00000280
IEF653I SUBSTITUTION JCL - SYSOUT=A,DCB=(RECFM=VBA,LRECL=137,BLKSIZE=7265),
XX SPACE=(CYL,(0,1)),UNIT=(DISK,3) 00000290
XXSYSUDUMP DD SYSOUT=A,SPACE=(CYL,(0,5)) 00000300
//GO,FT08F001 DD UNIT=(9TRACK,,DEFER),LABEL=(,BLP,,IN),DISP=OLD,
// DCB=(BLKSIZE=32000,RECFM=U,DEN=3),
// VOL=SER=JJ0097
//GO,DATA5 DD *

IEF236I ALLOC. FOR YZRWPMA2 GO
IEF237I 231 ALLLOCATED TO PGM=*,DD
IEF237I 232 ALLLOCATED TO FT05F001
IEF237I 333 ALLLOCATED TO FT06F001
IEF237I 333 ALLLOCATED TO FT07F001
IEF237I 331 ALLLOCATED TO SYSPRINT
IEF237I 231 ALLLOCATED TO SYSPRINT
IEF237I 232 ALLLOCATED TO SYSPRINT
IEF237I 333 ALLLOCATED TO SYSUDUMP
IEF237I 004 ALLLOCATED TO FT08F001

RECORD 1 OF FILE 2
LENGTH = 1800 BYTES

4B4BC1E3 E2D940F0 F4F0F3F0 F0F1F340 40404040 40404040 40404040 40404040 40404040 40404040
40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040 40404040
F0F0F2F3 F9F6F5F1 F1F9F1F9 F9F2F0F1 F1F7F1F7 F5F5F3F0 40F0F0F0 F0F2F3F9 F6F5F1F1 F9F1F9F9
F0404040 40404040 40404040 40404040 40404040 40404040 40F3F9F5 F940F0F0 F0F0F2F3 F9F6F5F1
F1F9F1F9 F9F061F6 F0F3F4F7 F3F0F0F0 40F0F0F0 F0F2F3F9 F6F5F1F1 F9F1F9F8 F5404040 40404040
40404040 40404040 40404040 40404040 60F8F6F7 F940F0F0 F0F0F2F3 F9F5F5F1 F1F9F2F0 F0E47FF1
F1F7F1F7 F5F6F3F4 40F0F0F0 F0F2F3F9 F5F5F1F1 F9F2F0F0 F2404040 40404040 40404040 40404040
40404040 40404040 40F3F9F5 F940F0F0 F0F0F2F3 F9F5F5F1 F1F9F2F0 F0F261F6 F0F3F4F7 F3F0F0F0
40F0F0F0 F0F2F3F9 F6F5F1F1 F9F1F9F9 F6404040 40404040 40404040 40404040 40404040 40404040
60F8F6F7 F940F0F0 F0F0F2F3 F9F6F5F1 F1F9F2F0 F0F17FF1 F1F7F1F7 F5F6F3F8 40F0F0F0 F0F2F3F9
F5F5F1F1 F9F1F9F9 F2404040 40404040 40404040 40404040 40404040 40404040 40F3F9F5 F940F0F0
F0F0F2F3 F9F6F5F1 F1F9F2F0 F1F461F6 F0F3F4F7 F3F0F0F0 40F0F0F0 F0F2F3F9 F6F5F1F1 F9F1F9F8
F5404040 40404040 40404040 40404040 40404040 40404040 60F8F6F7 F940F0F0 F0F0F2F3 F9F6F5F1
F1F9F1F9 F8F07FF1 F1F7F1F7 F5F6F4F2 40F0F0F0 F0F2F3F9 F5F5F1F1 F9F1F9F9 F6404040 40404040
40404040 40404040 40404040 40404040 40F3F9F5 F940F0F0 F0F0F2F3 F9F5F5F1 F1F9F2F0 F1F061F6
F0F3F4F7 F3F0F0F0 40F0F0F0 F0F2F3F9 F6F5F1F1 F9F1F9F8 F0404040 40404040 40404040 40404040
40404040 40404040 60F8F6F7 F940F0F0 F0F0F2F3 F9F5F5F1 F1F9F2F0 F0F37FF1 F1F7F1F7 F5F6F4F6
40F0F0F0 F0F2F3F9 F5F5F1F1 F9F1F9F9 F2404040 40404040 40404040 40404040 40404040 40404040
40F3F9F5 F940F0F0 F0F0F2F3 F9F5F5F1 F1F9F1F9 F8F251F6 F0F3F4F7 F3F0F0F0 40F0F0F0 F0F2F3F9
F6F5F1F1 F9F1F9F9 F3404040 40404040 40404040 40404040 40404040 40404040 60F8F6F7 F940F0F0